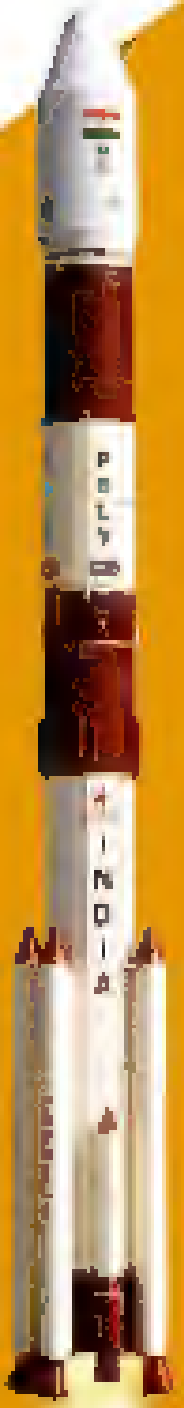
इसरो isro

PSLV-C26/IRNSS-1C



PSLV-C26

PSLV-C26 at First Launch Pad

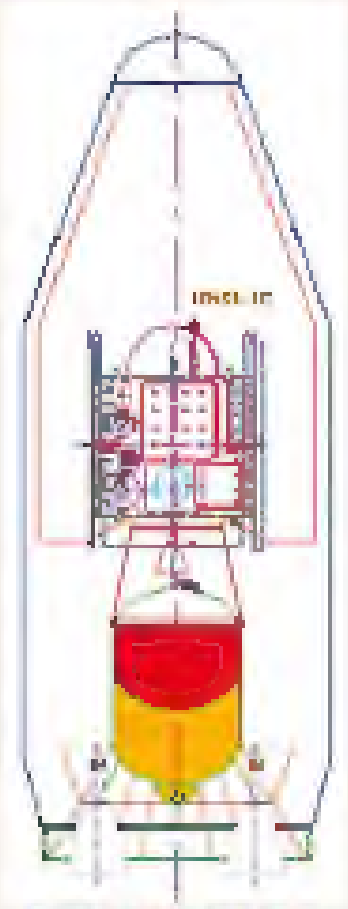
Polar Satellite Launch Vehicle, in its twenty eighth flight (PSLV-C26), will launch IRNSS-1C, the third satellite of the Indian Regional Navigation Satellite System (IRNSS). The launch will take place from the First Launch Pad (FLP) of Satish Dhawan Space Centre (SDSC) SHAR, Sriharikota. PSLV-C26 will use 'XL' version of PSLV. This is the seventh time 'XL' configuration is being flown, earlier six being PSLV-C11/Chandrayaan-1, PSLV-C17/GSAT-12, PSLV-C19/RISAT-1, PSLV-C22/IRNSS-1A, PSLV-C25/Mars Orbiter Spacecraft and PSLV-C24/IRNSS-1B missions.

PSLV-C26 at a glance (Vehicle Lift-off Mass 320 tonne Height 44.4 m)

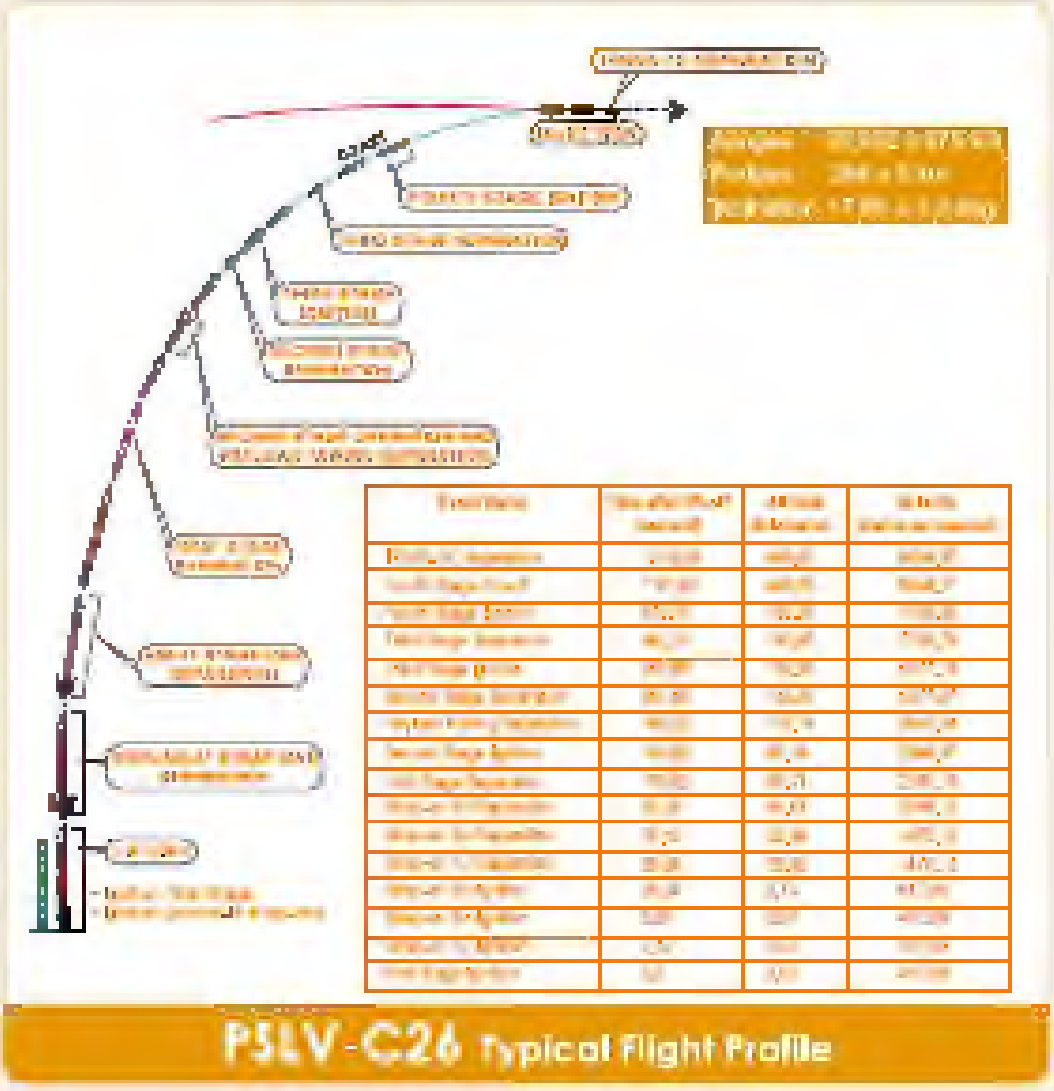
	Stage-I	Stage-II	Stage-III	Stage-IV
Nomenclature	Core Stage PS1 + 6 Strap-on Motors	PS2	PS3	PS4
Propellant	Solid (HTPB based)	Liquid ($UH25 + N_2O_4$)	Solid (HTPB based)	Liquid (MMH + MON3)
Mass (T)	138.2 (Core), 6 x 12.2 (Strap-on)	42.0	7.6	2.5
Max Thrust (kN)	4519 (Core), 6 x 716 (Strap-on)	804	240	7.3 x 2
Stage Dia (m)	2.8 (Core), 1 (Strap-on)	2.8	2.0	2.8
Stage Length (m)	20 (Core), 12 (Strap-on)	12.8	3.6	3.0

HTPB : Hydroxyl Terminated Poly Butadiene
UH25 : Unsymmetrical Dimethyl Hydrazine + 25% Hydrazine Hydrate
N_2O_4 : Nitrogen Tetroxide
MMH : Mono Methyl Hydrazine, MON-3: Mixed Oxides of Nitrogen

PSLV-C26



IRNSS-1C in PSLV-C26 Envelope



PSLV-C26 Typical Flight Profile

Nozzle End Segment of PSLV-C26 being hoisted for assembly

Assembly of PSLV-C26 Third and Fourth Stages

IRNSS-1C

IRNSS-1C is the third navigation satellite of the seven satellites constituting the IRNSS space segment. Its predecessors, IRNSS-1A and IRNSS-1B were launched by PSLV-C22 and PSLV-C24 in July 2013 and April 2014 respectively. IRNSS-1C has a lift-off mass of 1425.4 kg. The configuration of IRNSS-1C is similar to that of IRNSS-1A and IRNSS-1B. The satellite has been realised in less than six months after the launch of its predecessor.

The two solar panels of IRNSS-1C consisting of Ultra Triple Junction solar cells generate about 1660 Watts of electrical power. Sun and Star sensors as well as gyroscopes provide orientation reference for the satellite. Special thermal control schemes have been designed and implemented for some of the critical elements such as atomic clocks. The Attitude and Orbit Control System (AOCS) of IRNSS-1C maintains the satellite's orientation with the help of reaction wheels, magnetic torquers and thrusters. Its propulsion system consists of a Liquid Apogee Motor (LAM) and thrusters.

IRNSS-1C spacecraft undergoing Electro-Magnetic Interference and Electro-Magnetic Compatibility (EMI/EMC) tests

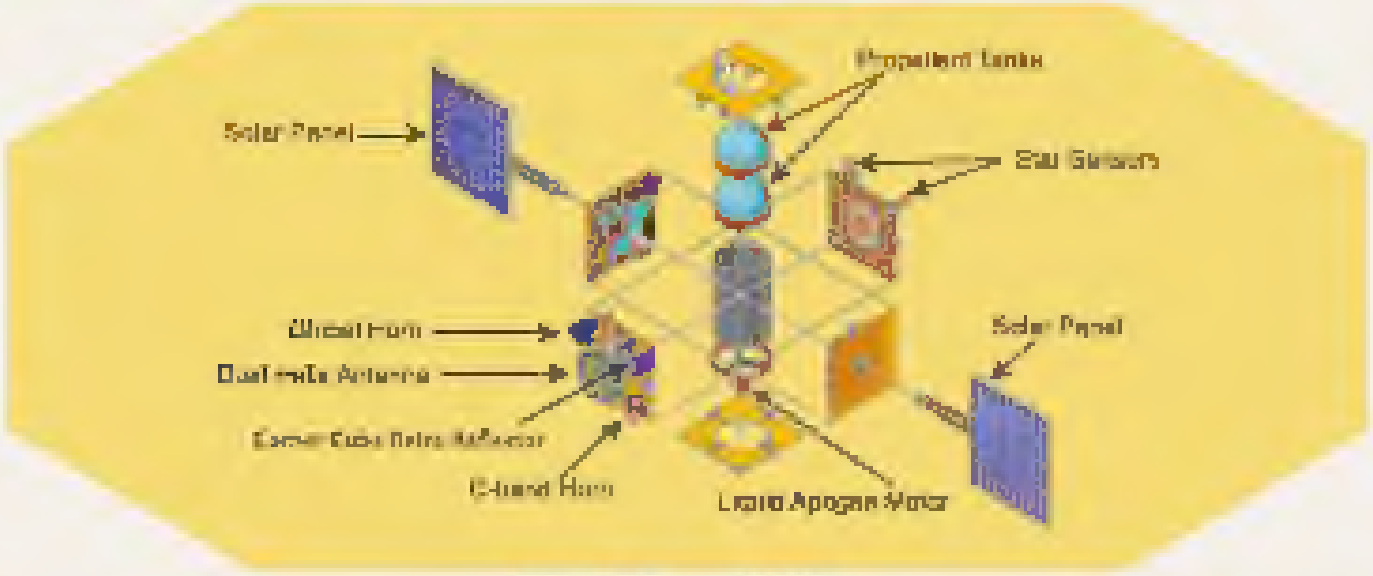
IRNSS-1C will be launched into a sub Geosynchronous Transfer Orbit (sub GTO) with a 284 km perigee (nearest point to Earth) and 20,650 km apogee (farthest point to Earth) with an inclination of 17.86 deg with respect to the equatorial plane.

After injection into this preliminary orbit, the two solar panels of IRNSS-1C are automatically deployed in quick succession and the Master Control Facility (MCF) at Hassan takes control of the satellite and performs the initial orbit raising manoeuvres consisting of one manoeuvre at perigee (nearest point to earth) and three at apogee (farthest point to earth). For these manoeuvres, the Liquid Apogee Motor (LAM) of the satellite is used, thereby finally placing it in the circular geostationary orbit at its designated location.

IRNSS-1C spacecraft being assembled with PSLV-C26

IRNSS-1C Salient features

ORBIT	Geostationary, at 83 deg East longitude
LIFT-OFF MASS	1425.4 kg
DRY MASS	600.1 kg
PHYSICAL DIMENSIONS	1.58 metre x 1.50 metre x 1.30 metre
POWER	Two solar panels generating 1660 W, one Lithium-Ion battery of 90 Ampere-Hour capacity
PROPULSION	440 Newton Liquid Apogee Motor, twelve 22 Newton thrusters
CONTROL SYSTEM	Zero momentum system, attitude input from Sun & Star Sensor and Gyroscopes. Reaction Wheels, Magnetic Torquers and 22 Newton thrusters as actuators
MISSION LIFE	10 years



IRNSS-1C Deployed View

PAYLOADS:

IRNSS-1C carries two types of payloads – navigation payload and ranging payload. The navigation payload of IRNSS-1C will transmit navigation service signals to the users. This payload will be operating in L5 band (1176.45 MHz) and S band (2492.028 MHz). A highly accurate Rubidium atomic clock is part of the navigation payload of the satellite. The ranging payload of IRNSS-1C consists of a C band transponder which facilitates accurate determination of the range of the satellite. IRNSS-1C also carries Corner Cube Retro Reflectors for laser ranging.

IRNSS Overview:

IRNSS is an independent regional navigation satellite system being developed by India. It is designed to provide accurate position information service to users in India as well as the region extending up to 1500 km from its boundary, which is the primary service area of IRNSS. The Extended Service Area lies between primary service area and area enclosed by the rectangle from Latitude 30 deg South to 50 deg North, Longitude 30 deg East to 130 deg East.

IRNSS will provide two types of services, namely, Standard Positioning Service (SPS) which is provided to all the users and Restricted Service (RS), which is an encrypted service provided only to the authorised users. The IRNSS System is expected to provide a position accuracy of better than 20 m in the primary service area.

IRNSS comprises of a space segment and a ground segment. The IRNSS space segment consists of seven satellites, with three satellites in geostationary orbit and four satellites in inclined geosynchronous orbit. IRNSS-1A and IRNSS-1B, the first two satellites of the IRNSS constellation, have already started functioning from their designated orbital slot after extensive co-orbit testing and evaluation to confirm their satisfactory performance.

IRNSS ground segment is responsible for navigation parameter generation and transmission, satellite control, ranging and integrity monitoring as well as timekeeping.

The constituent elements of the IRNSS ground segment are:

- ISRO Navigation Centre (INC) at Byalalu, is the nerve center of the IRNSS Ground Segment. INC primarily generates navigation parameters.
- IRNSS Range and Integrity Monitoring Stations (IRIMS) perform continuous one way ranging of the IRNSS satellites and are also used for integrity determination of the IRNSS constellation.
- IRNSS CDMA Ranging Stations (IRCDR) carry out precise two way ranging of IRNSS satellites.
- IRNSS Network Timing Centre (IRNWT) at Byalalu generates, maintains and distributes IRNSS Network Time.
- Spacecraft Control Facility (SCF) controls the space segment through Telemetry Tracking & Command networks. In addition to the regular TT&C operations, IRSCF also uploads the navigation parameters generated by the INC.

IRNSS Data Communication Network (IRDCN) provides the required digital communication backbone to IRNSS network.

International Laser Ranging Stations (ILRS) is being used periodically to calibrate the IRNSS orbit determined by the other techniques.

Applications of IRNSS:

- Terrestrial, Aerial and Marine Navigation
- Vehicle tracking and fleet management
- Precise Timing
- Terrestrial navigation aid for hikers and travellers
- Disaster Management
- Integration with mobile phones
- Mapping and Geodetic data capture
- Visual and voice navigation for drivers

Indian Space Research Organisation

Publications and Public Relations,
ISRO Headquarters, Antariksh Bhavan
New BEL Road, Bangalore - 560 231, India
www.isro.gov.in